

IN THE CLAIMS:

The following listing of claims is provided as a courtesy, as no amendments have been made.

Listing of Claims:

Claims 1 to 16 (cancelled).

Claim 17 (previously presented): A method for manufacturing vane segments for a gas turbine comprising the steps of:

providing a plurality of vanes,

manufacturing a vane segment from the plurality of vanes via powder metallurgy injection molding, the step of manufacturing including the steps of:

mixing a metal powder having a binding agent to form a homogeneous material, the metal powder accounting for at least 50% of the homogeneous material;

forming at least one molded body from the homogeneous material via injection molding,

subjecting the at least one molded body to a debinding process, and

compressing the at least one molded body via sintering to form the vane segment.

Claim 18 (previously presented): The method as recited in Claim 17, wherein the vane segment is designed as a guide vane segment and includes at least two guide vanes.

Claim 19 (previously presented): The method as recited in Claim 18, wherein the guide vane segment includes three or four guide vanes.

Claim 20 (canceled).

Claim 21 (previously presented): The method as recited in Claim 17, wherein, in that in order to manufacture one vane segment from at least two vanes,

a molded body for each vane is manufactured separately via injection molding, and

the molded bodies are joined together prior to the debinding process to form one molded body for the vane segment.

Claim 22 (previously presented): The method as recited in Claim 21, wherein the molded bodies for the vanes are joined together prior to the debinding process in the green state to form one molded body for the vane segment.

Claim 23 (previously presented): The method as recited in Claim 21, wherein the one molded body for the vane segment is then subjected to a uniform debinding process and uniform sintering in the debinding and sintering steps.

Claim 24 (previously presented): The method as recited in Claim 17, wherein, in that to manufacture one vane segment from at least two vanes:

a molded body for each vane is manufactured separately via injection molding,
the molded bodies for the vanes undergo separate debinding processes, and
the molded bodies for the vanes are subsequently joined together to form one molded body for the vane segment.

Claim 25 (previously presented): The method as recited in Claim 24, wherein the molded bodies for the vanes are joined together in a presintered state to form one molded body for the vane segment.

Claim 26 (previously presented): The method as recited in Claim 24, wherein the one molded body for the vane segment is then subjected to uniform sintering in the sintering step.

Claim 27 (previously presented): The method as recited in Claim 17, wherein, to manufacture one vane segment from at least two vanes, a joint molded body for all vanes of the vane segment is manufactured via injection molding.

Claim 28 (previously presented): The method as recited in Claim 27, wherein the joint molded body for the vane segment is subjected to a uniform debinding process and uniform sintering.

Claim 29 (previously presented): A component for a gas turbine, comprising a guide vane segment manufactured from a plurality of guide vanes via powder metallurgy injection molding from a homogeneous material comprising at least 50% metal powder.

Claim 30 (canceled).

Claim 31 (previously presented): The component as recited in Claim 29, wherein the guide vane segment includes three or four guide vanes.

Claim 32 (previously presented): The method as recited in claim 29 wherein the plurality of guide vanes are connected via an inner cover band and an outer cover band.

Claim 33 (previously presented): A method for manufacturing guide vane segments for a gas turbine comprising the steps of:

providing a plurality of guide vanes,
manufacturing a guide vane segment from the plurality of vanes via powder metallurgy injection molding from a homogenous material comprising at least 50% metal powder.

Claim 34 (previously presented): The method as recited in claim 33 wherein the metal powder is selected from one of a titanium alloy and nickel alloy.

Claim 35 (previously presented): The method as recited in claim 29 wherein the metal powder is selected from one of a titanium alloy and nickel alloy.

Claim 36 (previously presented): The method as recited in claim 17 wherein the metal powder is selected from one of a titanium alloy and nickel alloy.

Claim 37 (previously presented): The method as recited in claim 33 wherein the metal powder comprises 50 to 70% of the homogenous material.

Claim 38 (previously presented): The method as recited in claim 29 wherein the metal powder comprises 50 to 70% of the homogenous material.

Claim 39 (previously presented): The method as recited in claim 17 wherein the metal powder comprises 50 to 70% of the homogenous material.

Claim 40 (previously presented): The method as recited in claim 33 wherein the gas turbine is part of an aircraft engine.

Claim 41 (previously presented): The method as recited in claim 29 wherein the gas turbine is part of an aircraft engine.

Claim 42 (previously presented): The method as recited in claim 17 wherein the gas turbine is part of an aircraft engine.